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Kaplan received his MS in Aeronautics
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Satellite Orbital Maneuvers | Identify Required Orbit.

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Spacecraft Dynamics and Control

2 G. Avanzini Spacecraft Attitude

Dynamics and Control $\tilde{v} =$

$(e_{1,1}x + e_{1,2}y + e_{1,3}z)E^1 + +$

$(e_{2,1}x + e_{2,2}y + e_{2,3}z)E^2 + +$

$(e_{3,1}x + e_{3,2}y + e_{3,3}z)E^3$ This means

that the components of \tilde{v} in FI can be

expressed as a function of those in FB as

follows: $X = e_{1,1}x + e_{1,2}y + e_{1,3}z$

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$$Y = e^{2,1}x + e^{2,2}y + e^{2,3}z$$

$Z = e^{3,1}x + e^{3,2}y + e^{3,3}z$ or, in compact matrix form, $v_I = LIBv_B$ where the transformation matrix LIB is given by

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Spacecraft Dynamics and
Control(1976),and Wiesel'sSpace' ight
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